



Lufttüchtigkeitsanweisung (LTA) Consigne de Navigabilité (CN) Direttive sulla Navigabilità (DN) Airworthiness Directive (AD)		FOCA AD <b>HB-2016-004</b>
Inkraftsetzung Mise en vigueur Entrata in vigore Effective Date	<b>17 May 2016</b>	<b>Pilatus – PC-9(M)</b> FOCA TC/TCDS No: F 56-32

<b>Issue Date:</b>	03 May 2016
<b>ATA Chapter:</b>	ATA 53 – Fuselage
<b>Subject:</b>	<b>Fuselage – Main Frames – Frame 11 Fittings – Inspection</b>
<b>Supersedure / Revised AD(s):</b>	Not applicable
<b>Type Certificate Holder's Name:</b>	Pilatus Aircraft Ltd.
<b>Manufacturer(s):</b>	Pilatus Aircraft Ltd.
<b>Applicability:</b>	Model PC-9(M) aircraft, Manufacturers Serial Numbers (MSN) 605 and MSN 617 through MSN 670.
<b>Reason:</b>	<p>This Airworthiness Directive (AD) is prompted due to a report of Stress Corrosion Cracking (SCC) on the Main Frame on Frame (FR) 11 left fitting Part Number (P/N) 112.35.07.489 and right fitting P/N 112.35.07.490.</p> <p>Such a condition, if left uncorrected, could lead to potential loss of the horizontal stabilizer.</p> <p>In order to correct and control the situation, this AD requires a one-time check to identify the material specification and inspect the affected areas of the airframe that are made of aluminium alloy AA2024-T351. Any structural parts of the aircraft structure found to be cracked must be reported to Pilatus prior to further flight.</p>
<b>Required Action(s) and Compliance Time(s):</b>	<p>Required as indicated below, unless already accomplished:</p> <ol style="list-style-type: none"> <li>(1) Within the next 120 days after the effective date of this AD, perform a test to identify the material specification of the left and right fittings P/N 112.35.07.489 and P/N 112.35.07.490 as required by paragraph (§) 3.B. of PILATUS PC-9(M) Service Bulletin No. 53-018.</li> <li>(2) If during the inspection required by § (1) of this AD, fittings made of aluminium alloy AA2124-T851 are found, report the inspection results as required by § 3.D. of PILATUS PC-9(M) Service Bulletin No. 53-018.</li> <li>(3) If during the inspection required by § (1) of this AD, fittings made of aluminium alloy AA2024-T351 are found, before next flight after the effective date of this AD, perform an inspection for cracks as required by § 3.C. of PILATUS PC-9(M) Service Bulletin No. 53-018 for cracks. No cracks are allowed.</li> </ol>

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**Note 1:** A scheduled maintenance interval of 12 months to check for cracks at the FR 11 fittings made of aluminium alloy AA2024-T351 will be added to Chapter 5 of the Aircraft Maintenance Manual (AMM).

- (4) If during the inspection required by § (3) of this AD, cracks are found, prior further flight, replace the fittings in accordance with the instructions given in PILATUS PC-9(M) Service Bulletin No. 53-019.

**Ref. Publication(s):**

PILATUS PC-9(M) Service Bulletin No. 53-018, initial issue.  
PILATUS PC-9(M) Service Bulletin No. 53-019, initial issue.

The use of later approved revisions of this document is acceptable for compliance with the requirements of this AD.

**For further information contact:**

The applicable manufacturer's documentation may be obtained directly from:

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Customer Technical Support (MCC)  
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