



Swiss Confederation

Lufttüchtigkeitsanweisung (LTA) Consigne de Navigabilité (CN) Prescrizione di Aeronavigabilità (PA) Airworthiness Directive (AD)		FOCA AD HB-2006-272
Inkraftsetzung Mise en vigueur Entrata in vigore Effective Date	15 June 2006	Pilatus – PC-7 MkII FOCA TC/TCDS No: F56-25 ATA 53 – FUSELAGE

Issue date: 08 June 2006

Subject: FUSELAGE STRUCTURE – FRAMES 3C – Inspection, Part replacement

Superseded / Revised ADs: Not applicable

1. APPLICABILITY:

PILATUS AIRCRAFT LTD.

All Model PC-7 MkII airplanes Manufacturer Serial Number (MSN) 101 thru MSN 616 inclusive.

2. REASON:

This Airworthiness Directive (AD) is prompted due to the discovery of Frame 3C foot fitting corrosion. Frame 3C is located on the fuselage structure in the pick-up brackets Part Number (P/N) 553.20.09.039 of Frame 3C.

Electrolytic action between the dissimilar metals of the bolts P/N 932.19.60.337 and the brackets P/N 553.20.09.039 has caused the corrosion. It is thought that such corrosion will predominantly occur on aircraft flying in a coastal environment. The Frame 3C fitting forms the fuselage component of the wing rear spar to fuselage joint. Failure of this component will prejudice the structural integrity of the airplane.

In order to correct and control the situation, this AD requires a one time inspection of the Frame 3C assemblies and adjacent structure for signs of corrosion and the replacement of the wing rear attachment bolt assemblies with cadmium plated bolts to prevent further corrosion. If the damage is out of limits, the fitting assembly and the attaching parts have to be replaced.

3. COMPLIANCE / ACTION:

Required as indicated below, unless already accomplished:

Within the next 150 flying hours time-in-service (TIS) or six (6) months after the effective date of this AD, whichever occurs first:

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3.1. INSPECTION / REPLACEMENT OF THE BOLT ASSEMBLIES

- 3.1.1. Perform an inspection of the Frame 3C for corrosion damage in the foot fittings and the surrounding area in accordance with paragraph 3. B. of PILATUS PC-7 MkII Service Bulletin No. 53-006. Corrosion is not permitted in the bolt holes. Surface corrosion on the bracket, shear plate on Frame 3C assembly is permitted if in limits of Structural Repair Manual (SRM) reference 53-20-00, Page Block 101. All other types of corrosion in the bracket, shear plate, Frame 3C or in the adjacent structure are not permitted.
- 3.1.2. If during the inspection required by paragraph 3.1.1. of this AD any corrosion is found, prior to further flight, remove the surface corrosion in accordance with paragraph 3. B. (4) of PILATUS PC-7 MkII Service Bulletin No. 53-006.
- 3.1.3. If during the action required by paragraph 3.1.2. of this AD any corrosion is out of the limits referenced in SRM, 53-20-00, Page Block 101, or if it is not possible to remove the surface corrosion and stay in limits, prior to further flight, replace the bracket, shear plate and Frame 3C assembly together.
- 3.1.4. If during the inspection required by paragraph 3.1.1. of this AD any corrosion in the skin or structure adjacent to Frame 3C is found, contact Pilatus Aircraft Ltd. for a repair solution.
- 3.1.5. Replace the bolt, washers and nut in accordance with paragraph 3. B. (5), (6) and (7) of PILATUS PC-7 MkII Service Bulletin No. 53-006.

4. REF. PUBLICATIONS:

The actions required by this AD shall be done in accordance with the manufacturer's documentation listed in this paragraph, and/or later revisions approved by the Swiss Federal Office of Civil Aviation (FOCA):

Manufacturer's Documentation

- PILATUS PC-7 MkII Service Bulletin No. 53-006, dated 14 February 2006.
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The applicable manufacturer's documentation may be obtained directly from:

PILATUS AIRCRAFT LTD.
CUSTOMER LIAISON MANAGER
CH-6371 STANS, Switzerland

TEL No.: +41 41 619 6226
FAX No.: +41 41 619 6170
Email: snolan@pilatus-aircraft.com

5. FOR FURTHER INFORMATION CONTACT:

FEDERAL OFFICE OF CIVIL AVIATION (FOCA)

Design and Production (STEH)
CH-3003 Bern, Switzerland

FAX No.: +41 31 325 9324 (or)
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