



Airworthiness Directive

AD No.: 2018-0217

Issued: 10 October 2018

Note: This Airworthiness Directive (AD) is issued by EASA, acting in accordance with Regulation (EU) 2018/1139 on behalf of the European Union, its Member States and of the European third countries that participate in the activities of EASA under Article 129 of that Regulation.

This AD is issued in accordance with Regulation (EU) 748/2012, Part 21.A.3B. In accordance with Regulation (EU) 1321/2014 Annex I, Part M.A.301, the continuing airworthiness of an aircraft shall be ensured by accomplishing any applicable ADs. Consequently, no person may operate an aircraft to which an AD applies, except in accordance with the requirements of that AD, unless otherwise specified by the Agency [Regulation (EU) 1321/2014 Annex I, Part M.A.303] or agreed with the Authority of the State of Registry [Regulation (EU) 2018/1139, Article 71 exemption].

Design Approval Holder's Name:

PILATUS AIRCRAFT Ltd

Type/Model designation(s):

PC-6 aeroplanes

Effective Date: 24 October 2018

TCDS Number(s): Switzerland No. F 56-10

Foreign AD: Not applicable

Supersedure: None

ATA 55 – Stabilizers – Horizontal Stabilizer Hinge Bracket Assembly / Front Spar Attachment Area – Inspection

Manufacturer(s):

Pilatus Aircraft Ltd; and Fairchild Republic Company, formerly Fairchild Industries, Fairchild Heli Porter and Fairchild-Hiller Corporation

Applicability:

PC-6 aeroplanes, all models, all manufacturer serial numbers.

Definitions:

For the purpose of this AD, the following definitions apply:

The SB: Pilatus Aircraft Ltd Service Bulletin (SB) No. 55-004.

Affected part: Left Hand (LH) and Right Hand (RH) hinge bracket assemblies, having Part Number (P/N) 6304.0021.51 and P/N 6304.0021.52; and screws (pan head), having P/N 931.54.41.753.

Serviceable part: An affected part that is new, or has passed an inspection (no defects found), before installation, in accordance with the instructions of Section 3.C of the SB, or has been corrected in accordance with the instructions of Section 4 of the SB.



Reason:

During a routine inspection, the rivets of the hinge bracket assemblies on a Pilatus PC-6 were found to be sheared or missing. Investigation results identified that this was most likely due to application of too much force to the ends of the horizontal stabilizer during ground handling.

This condition, if not detected and corrected, could lead to failure of the primary horizontal stabilizer load path and consequent separation of the horizontal stabilizer, possibly resulting in loss of control of the aeroplane.

To address this potential unsafe condition, Pilatus Aircraft Ltd issued the SB to provide applicable inspection instructions.

For the reasons described above, this AD requires a one-time inspection of the affected parts and the horizontal stabilizer front spar attachment area and, depending on findings, accomplishment of applicable corrective action(s). This AD also requires, before installation, inspection of, and, depending on findings, corrective action(s) on, affected parts held as spare.

Required Action(s) and Compliance Time(s):

Required as indicated, unless accomplished previously:

Inspection(s):

- (1) During the next scheduled maintenance (100 flight hours or Annual Inspection), or within 12 months after the effective date of this AD, whichever occurs first, inspect the affected parts and the horizontal stabilizer front spar attachment area in accordance with the instructions of Section 3.C of the SB.

Corrective Action(s):

- (2) If, during the inspection as required by paragraph (1) of this AD, any discrepancies are found, as defined in the SB, before next flight, accomplish the applicable corrective action(s) in accordance with the instructions of Section 3 of the SB.
- (3) If, during the inspection as required by paragraph (1) of this AD, any damage is found on the front spar of the horizontal stabilizer that cannot be removed by installation of oversize rivets, before next flight, contact Pilatus Aircraft Ltd for approved repair instructions and accomplish those instructions accordingly.

Parts Installation:

- (4) From the effective date of this AD, it is allowed to install on any aeroplane an affected part, provided it is a serviceable part, as defined in this AD.

Ref. Publications:

Pilatus Aircraft Ltd SB No. 55-004 original issue dated 02 July 2018.

The use of later approved revisions of the above-mentioned document is acceptable for compliance with the requirements of this AD.



Remarks:

1. If requested and appropriately substantiated, EASA can approve Alternative Methods of Compliance for this AD.
2. This AD was posted on 07 September 2018 as PAD 18-125 for consultation until 05 October 2018. No comments were received during the consultation period.
2. Enquiries regarding this AD should be referred to the EASA Safety Information Section, Certification Directorate. E-mail: ADs@easa.europa.eu.
3. Information about any failures, malfunctions, defects or other occurrences, which may be similar to the unsafe condition addressed by this AD, and which may occur, or have occurred on a product, part or appliance not affected by this AD, can be reported to the [EU aviation safety reporting system](#).
4. For any question concerning the technical content of the requirements in this AD, please contact: Pilatus Aircraft Ltd, Customer Support Manager, CH-6371 Stans, Switzerland
Telephone: +41 41 619 33 33, Fax: +41 41 619 73 11
E-mail: SupportPC12@pilatus-aircraft.com, Website: www.pilatus-aircraft.com.

