Service Bulletin No:	**24-005**	**Ref No:**	**181**
Modification No:	**EC-18-0038**	**ATA Chapter:**	**24**

ELECTRICAL POWER - DC ELECTRICAL LOAD DISTRIBUTION
INTRODUCTION OF ADDITIONAL ATTACHMENT HARDWARE TO HOLD A CABLE LOOM AT THE LH COCKPIT RAIL INSTALLATION

1. Planning Information

A. Effectivity

MSN 155 thru 251, MSN 283 thru 290, MSN 293 thru 302.

This modification will be incorporated in production on MSN 252 thru 282, MSN 291 and 292 and MSN 303 and subsequent.

B. Concurrent Requirements

None.

C. Reason

(1) Problem

Five wires that are part of a cable loom behind the LH cockpit rail installation were found to be damaged. The damage was caused by the attachment screws of the relays K106 and K107. The relays are installed to the frame support assembly of the LH cockpit rail installation.

(2) Solution

The frame support assembly is partially removed and a visual inspection is made of the cable loom for chafing and damage. If necessary, the chafed and damaged wires are repaired. To prevent future chafing and damage to the wires, additional attachment hardware is installed to hold the cable loom away from the attachment screws of the relays K106 and K107.

D. Description

This Service Bulletin gives the data and instructions necessary to:

- Do a visual inspection of the cable loom behind the frame support assembly of the LH cockpit rail installation
- Install additional attachment hardware to the cable loom behind the frame support assembly of the LH cockpit rail assembly.

E. Compliance

Mandatory.

Accomplishment required not later than 90 days after the effective date of this Service Bulletin.

F. Approval

The technical content of this Service Bulletin is approved under the authority of Letter of DOA Acceptance ref. FOCA.21J.002.

Pilatus advises Operators/Owners to check with their designated Airworthiness Authorities for any changes, local regulations or sanctions that may affect the embodiment of this Service Bulletin.

G. Copyright

H. Manpower

	Man-Hours
Preparation	1.00
Modification	1.00
Close up	1.00
TOTAL MAN-HOURS	**3.00**

I. Weight and Balance

(1) Weight Change

Not changed.

(2) Moment Change

Not changed.

J. Electrical Load Data

Not changed.

K. Software

Not changed.

L. References

Aircraft Maintenance Manual (AMM), 24-00-00-00A-012A-A, 24-00-00-00A-525A-A, 25-10-00-00A-040A-A, 95-00-00-00A-012A-A, 95-10-00-00A-520A-A, 95-10-00-00A-720A-A.

M. Publications Affected

Illustrated Parts Data (IPD), 24-60-00-11E-941A-A, 24-60-00-11H-941A-A.

N. Interchangeability of Parts

Not applicable.

2. Material Information

A. Material - Price and Availability

Modification Kit No. 500.50.21.075 is necessary to do this Service Bulletin.

Operators who require further information on Price and Availability should contact their Customer Liaison Manager at:

Pilatus Aircraft Ltd,
6371 Stans,
Switzerland.

Operators are requested to advise Pilatus Aircraft Ltd. of the Manufacturer's Serial Number (MSN), the flying hours and landings of aircraft which are allocated for this Service Bulletin using the Service Bulletin Evaluation Form.

NOTE: When you order the modification kit (P/N 500.50.21.075) from Pilatus Aircraft Ltd, you will also get a parts list. Use the numbers in column 1 (pos.) to identify the parts in the kit (Ref. Para. 2.B.(1), Column 1).

B. Material Necessary for Each Aircraft

(1) Material to be Procured

Modification Kit No. 500.50.21.075.

The table below lists the parts in the Modification Kit (Ref. Para. 2.A.) and the disposition of the replaced parts:

POS. NO.	DESCRIPTION	OLD PART NO.	QTY	DISP. CODE	FIG. NO.	ITEM. NO.
10	BUSH, DISTANCE	-	1	N	1	9
20	MOUNTING BASE	-	1	N	1	8
30	SCREW, PAN HD	-	1	N	1	5
40	NUT	-	1	N	1	11
50	WASHER	-	1	N	1	10
Disposition Codes: D - Discard / N - New / R - Return to Pilatus						

(2) Operator Supplied Materials (Ref. AMM 00-50-00-00A-013A-A)

MATERIAL NO.	DESCRIPTION	QTY	REMARKS
P09-005	CABLE TIE	A/R	P/N 971.32.51.105 (Fig. 1, item 6)
P09-064	TAPE, SILICONE RED, FIBERGLASS	A/R	P/N 917.40.60.311 (Fig. 1, item 7)

C. Material Necessary for Each Spare

None.

D. Re-identified Parts

None.

E. Tooling - Cost and Availability

None.

3. Accomplishment Instructions

WARNING: READ AND OBEY THE SAFETY PRECAUTIONS AT THE START OF CHAPTER 95, CREW ESCAPE AND SAFETY, BEFORE YOU GO IN OR NEAR TO THE COCKPIT. IF THE EJECTION SEAT AND THE CANOPY FRACTURING SYSTEM (CFS) OPERATE ACCIDENTALLY OR INCORRECTLY THEY CAN CAUSE DEATH OR INJURY TO PERSONNEL AND/OR DAMAGE TO EQUIPMENT.

NOTE: For the safety precautions for the ejection seat and the CFS, refer to AMM, 95-00-00-00A-012A-A.

A. Preparation

(1) Do the safety procedures for the electrical system before you do the work to the electrical system, refer to AMM, 24-00-00-00A-012A-A.

(2) Remove the front ejection seat, refer to AMM, 95-10-00-00A-520A-A.

(3) Remove the front-cockpit lower-left forward panel, refer to AMM, 25-10-00-00A-040A-A.

B. Modification (Ref. Fig. 1)

(1) Identify the frame support assembly (4) on the LH cockpit rail installation (Ref. Detail A).

(2) Remove one screw (3) and washer (2) and then carefully remove the distance bush (1) from the frame support assembly (4).

NOTE: Take care to hold the bush (1) behind the frame support assembly (4) when you remove the screw (3) and washer (2).

(3) Remove the remaining three screws (3), washers (2) and distance bushes (1) in turn.

NOTE: Hold the frame support assembly (4) and take care to hold the bushes (1) when you remove each of the remaining screws (3) and washers (2).

(4) Carefully pull the frame support assembly (4) sufficiently away from the aircraft structure to get access to the rear of the frame support assembly (4).

(5) Do a visual inspection of the cable loom behind the frame support assembly (4).

(6) If you find damage to the cable loom, contact Pilatus for technical support and do the repairs as necessary. When you have done the repairs, or if you do not find damage, continue with this Service Bulletin from Step 3.B.(7).

(7) Put the new attachment hardware in position on the rear side of the frame support assembly (4) as follows:

- The screw (5) (Pos. No. 30)
- The mounting base (8) (Pos. No. 20)
- The distance bush (9) (Pos. No. 10).

(8) Loosely install the new washer (10) (Pos. No. 50) and the new nut (11) (Pos. No. 40) on the screw (5) (Pos. No. 30).

(9) Align the mounting base (8) (Pos. No. 20) longitudinally with the frame support assembly (4) and hold it in position.

(10) Tighten the nut (11) (Pos. No. 40) on the screw (5) (Pos. No. 30).

(11) In the area where the cable loom interfaces with the mounting base (8) (Pos. No. 20), apply the red fiberglass tape (7) (Material No. P09-064) to the cable loom (Ref Detail C).

(12) Install the new cable tie (6) (Material No. P09-005) to the mounting base (8) (Pos. No. 20) and the cable loom.

(13) Make sure that there is clearance between the cable loom and the attachment screws of the relays K106 and K107.

(14) Put the frame support assembly (4) in position and align it with the installation holes on the aircraft structure.

(15) Align one of the distance bushes (1) with an installation hole and carefully hold in position.

(16) Loosely install the screw (3) and washer (2) in the frame support assembly (4).

(17) Loosely install the remaining three screws (3), washers (2) and distance bushes (1) in turn.

(18) Tighten the four screws (3).

C. Close up

(1) Remove all tools and materials. Make sure the work areas are clean.

(2) Do the close up procedures for the electrical system, refer to AMM, 24-00-00-00A-525A-A.

(3) Install the front-cockpit lower-left forward panel, refer to AMM, 25-10-00-00A-040A-A.

(4) Install the front ejection seat, refer to AMM, 95-10-00-00A-720A-A.

D. Documentation

(1) Make an entry in the Aircraft Logbook that this Service Bulletin has been incorporated.

(2) Use the Service Bulletin Evaluation Sheet and report your results and the serial number of the aircraft to Pilatus.

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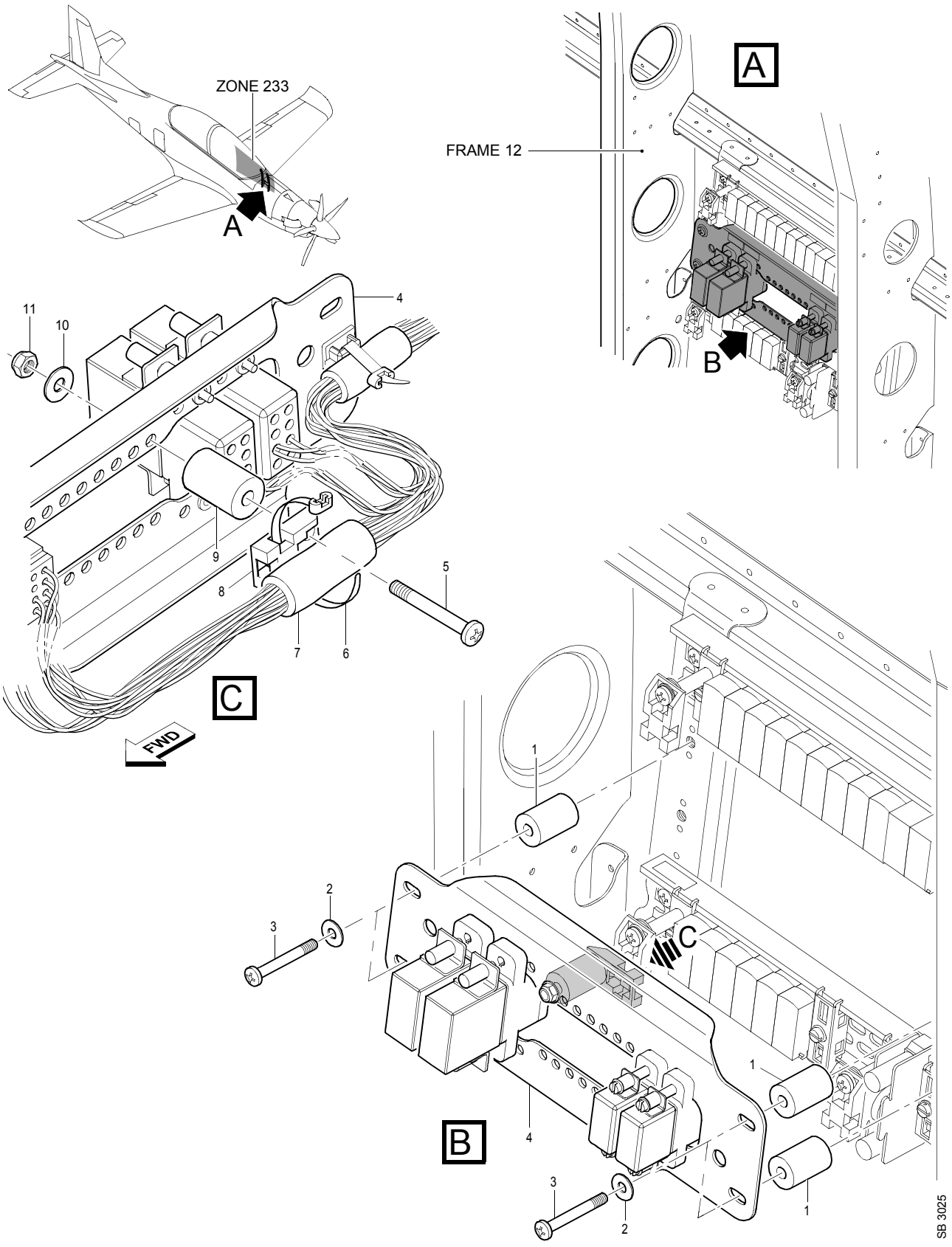

Installation of Additional Attachment Hardware to the LH Cockpit Rail Assembly Wiring-Loom
Figure 1

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SERVICE BULLETIN EVALUATION SHEET FOR SB No. 24-005	
Title	**Electrical Power - DC Electrical Load Distribution - Introduction of Additional Attachment Hardware to Hold a Cable Loom at the LH Cockpit Rail Installation**
Customer	
Service Center	

EMBODIMENT REPORTING		
This SB has been embodied:		**On the entire fleet**
		Only partially

Provide embodiment details per aircraft (use additional copies of this table, if necessary)

MSN	Flying Hours	MSN	Flying Hours

Additional embodiment comments/findings

EDITORIAL COMMENTS (procedure, kit quality, suggested improvements, etc.)

Name	Signature	Date

Please complete and forward this form to:

Pilatus Aircraft LTD,
Customer Technical Support (MCC),
P.O. BOX 992,
6371 Stans, Switzerland
Fax: +41 (0)41 619 6773
Email: Techsupport@pilatus-aircraft.com

SERVICE BULLETIN EVALUATION SHEET

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