

Federal Department of the Environment, Transport, Energy and Communications DETEC

Federal Office of Civil Aviation FOCA

Safety Division - Aircraft

Swiss Confederation

Lufttüchtigkeitsanweisung (LTA) Consigne de Navigabilité (CN) Direttive sulla Navigabilità (DN) Airworthiness Directive (AD)		FOCA AD HB-2012-001
Inkraftsetzung Mise en vigueur Entrata in vigore Effective Date	30 April 2012	Pilatus – PC-21 FOCA TC/TCDS No: F 56-35

Issue Date: 16 April 2012

ATA Chapter: ATA 24 – Electrical Power

Subject: Electrical Power – LH & RH Wire Looms In The Front Cockpit

Adjacent To Frame 10 - Inspection/Modification

Supersedure / Revised

AD(s):

Not applicable

Type Certificate Holder's Name:

Pilatus Aircraft Ltd.

Manufacturer(s): Pilatus Aircraft Ltd.

Applicability: Model PC-21 aircraft, Manufacturer Serial Numbers (MSN) 101 through

MSN 152 inclusive.

Reason: This Airworthiness Directive (AD) is prompted due to the discovery of

chafing between the front left cockpit wiring harness and the LH side panel next to the pedal. In addition burning marks were found in the area

due to a short circuit.

Such a condition, if left uncorrected, could lead to fire and smoke in the cockpit including partial loss of the wire harness, resulting in reduced

controllability of the aircraft.

In order to correct and control the situation, this AD requires a one-time check and additional, the installation of a protective sleeve to the left and

right wire harness.

Required Action(s) and Compliance Time(s):

Required as indicated below, unless already accomplished:

(1) Within the next 90 days after the effective date of this AD, inspect the LH and RH aircraft wire looms as required by paragraph (§) 3.B. of PILATUS PC-21 Service Bulletin No. 24-002 and modify the wire looms by installing an additional protection sleeve as requested by

(§) 3.C. of PILATUS PC-21 Service Bulletin No. 24-002.

Ref. Publication(s): PILATUS PC-21 Service Bulletin No. 24-002, initial issue.

The use of later approved revisions of this document is acceptable for

compliance with the requirements of this AD.

Legal base: Art. 26, 51 and 51A (Ordinance on the airworthiness of aircraft: SR 748.215.1)

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HB-2012-001

For further information contact:

The applicable manufacturer's documentation may be obtained directly from:

PILATUS AIRCRAFT LTD. Customer Technical Support (MCC) P.O. Box 992 CH-6371 Stans, Switzerland

Tel.: +41 (0)41 619 67 74 Fax: +41 (0)41 619 67 73

E-mail: <u>Techsupport@pilatus-aircraft.com</u>

For further information contact:

FEDERAL OFFICE OF CIVIL AVIATION (FOCA) Design and Production (STEH) CH-3003 Bern, Switzerland

Fax: +41 (0) 31 322 59 18 E-mail: airdir@bazl.admin.ch