

SERVICE BULLETIN

SERVICE BULLETIN NO: 30-009

REF NO: BS

MODIFICATION NO:

ATA CHAPTER: 30

ICE AND RAIN PROTECTION - WINDSHIELD DEICE SYSTEM WINDSHIELD DEICE SYSTEM WIRING - INSPECTION

1. Planning Information

A. Effectivity

PC-12 and PC-12/45 aircraft MSN 322 thru MSN 400.

B. Concurrent Requirements

None

C. Reason

(1) Problem

It is possible for heat damage to occur in given wires in the deice systems of the left and/or right windshields in some PC-12 and PC-12/45 aircraft.

(2) Cause

This problem has only occurred in a windshield deice system in which wires of incorrect gage were used.

(3) Solution:

- Do an inspection to make sure the wires installed in the windshield deice systems agree with the specifications given in the applicable wiring diagrams
- Replace all incorrect wires

D. Description

This Service Bulletin gives the data and instructions necessary to do an inspection of the left and right windshield deice system wires.

E. Compliance

Mandatory.

Required within the next 10 hours time-in-service but not later than three calendar months after the effective date of this Service Bulletin.


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F. Approval

The technical content of this Service Bulletin is approved by the Federal Office for Civil Aviation (FOCA) of Switzerland as an Airworthiness Directive.

PILATUS advises Operators/Owners to check with their local Airworthiness Authorities for any changes, local regulations or sanctions that may affect the embodiment of this Service Bulletin.

G. Manpower

	Total
Preparation (overhead panel not fully removed)	1.5
Inspection	1.0
Close up	1.5
TOTAL MAN-HOURS	4.0

H. Weight and Balance

(1) Weight Change

Not affected.

(2) Moment Change

Not affected.

I. Electrical Load Data

Not changed.

J. Software

Not changed.

K. References

Aircraft Maintenance Manual (AMM): 24-00-00, 24-30-07, 25-10-03 and 30-40-00.

Wiring Manual (WM): 30-40-00 and 30-40-10.

L. Publications Affected

Not applicable.

M. Interchangeability of Parts

Not applicable.


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2. Material Information

A. Material Necessary for Each Aircraft

(1) Material to be Purchased

A Modification Kit is not necessary for this Service Bulletin.

(2) Additional Material to be Procured

Not applicable.

(3) Operator Supplied Materials

Not applicable.

B. Material Necessary for Each Spare

Not applicable.

C. Reidentified Parts

Not applicable.

D. Tooling - Cost and Availability

Not applicable.

E. Warranty

Credit will be issued for all affected aircraft on the approval of a warranty claim provided the work is accomplished by an authorized Service Center within three months from the issue date of this Service Bulletin.

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3. Accomplishment Instructions - Aircraft

A. Preparation

- (1) Obey the safety precautions given in Electrical Power - Maintenance Practices (Ref. AMM, 24-00-00, Page Block 201).
- (2) Disconnect the battery (Ref. AMM, 24-30-07, Page Block 401).
- (3) Put warning signs (DO NOT APPLY ELECTRICAL POWER) in the flight compartment and on the external power receptacle.
- (4) Remove sufficient fasteners from the rear of the overhead panel in the flight-compartment for access to the area of terminal block (TB152) (Ref. AMM, 25-10-03, Page Block 401). If necessary remove related components as applicable to let you lower the panel.

B. Inspection (Ref. Fig 1 and 2)

Wire	Splice	Link Wire	Terminal Block			Link Wire	Splice	Wire
H110A10	SP6	AWG12	A	TB152-01	C	AWG12	SP10	H110C10
H107A10	SP7	AWG12	D	TB152-01	F	AWG12	SP11	H107C10
H251A10	SP8	AWG12	A	TB152-02	C	AWG12	SP12	H251C10
H250A10N	SP9	AWG12	D	TB152-02	F	AWG12	SP13	H250B10
H307A10	SP27	AWG12	A	TB152-08	C	AWG12	SP31	H307C10
H308A10	SP28	AWG12	D	TB152-08	F	AWG12	SP32	H308C10
H309A10	SP29	AWG12	A	TB152-09	C	AWG12	SP33	H309C10
H310A10N	SP30	AWG12	D	TB152-09	F	AWG12	SP34	H310B10

Wire, Splice and Link Wire Assemblies - Inspection

- (1) Do an inspection of the wires, splices and related link wires in the area of terminal block (TB152) for signs of heat damage (Ref. Table - Wire, Splice and Link Wire Assemblies - Inspection). Heat damage is not permitted. Replace all damaged wires and link wires (Ref. Para C).
- (2) Do a check of the wires and related link wires to make sure that they are correct as specified (Ref. Table - Wire, Splice and Link Wire Assemblies - Inspection and WM, 30-40-00 and 30-40-10).

C. Replacement

Steps (1) thru (7) are only applicable if you found damaged, or incorrect wires during the inspection procedure (Ref. Para B).

- (1) Fully remove the flight-compartment overhead panel for access to terminal block TB152 (Ref. AMM, 25-10-03, Page Block 401).
- (2) Replace damaged or incorrect wires as necessary (Ref. WM, 30-40-00 and 30-40-10).

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- (3) Open and safety the circuit breakers:

DOME LIGHT 100% (BATTERY BUS)
COCKPIT FLOOD LIGHT (BATTERY BUS)
AUDIO (BATTERY BUS)
AUDIO (AVIONIC 2 BUS)

- (4) Remove the applicable warning signs from the flight compartment and the external power receptacle.

- (5) Energize the aircraft electrical system (Ref. AMM, 24-00-00, Page Block 201)

- (6) Do the Windshield De-ice System Adjustment/Test procedure (Ref. AMM, 30-40-00 (CONFIG 2), Page Block 501).

- (7) Open and safety the circuit breaker:

WSHLD DE-ICE (GENERATOR 2 BUS)

D. Close up

- (1) Remove all tools and materials. Make sure the work areas are clean.

- (2) Install the flight-compartment overhead panel (Ref. AMM, 25-10-03, Page Block 401).

- (3) If applicable, remove the applicable warning signs from the flight compartment and the external power receptacle.

- (4) If applicable, Close circuit breakers:

DOME LIGHT 100% (BATTERY BUS)
COCKPIT FLOOD LIGHT (BATTERY BUS)
AUDIO (BATTERY BUS)
AUDIO (AVIONIC 2 BUS)
WSHLD DE-ICE (GENERATOR 2 BUS)

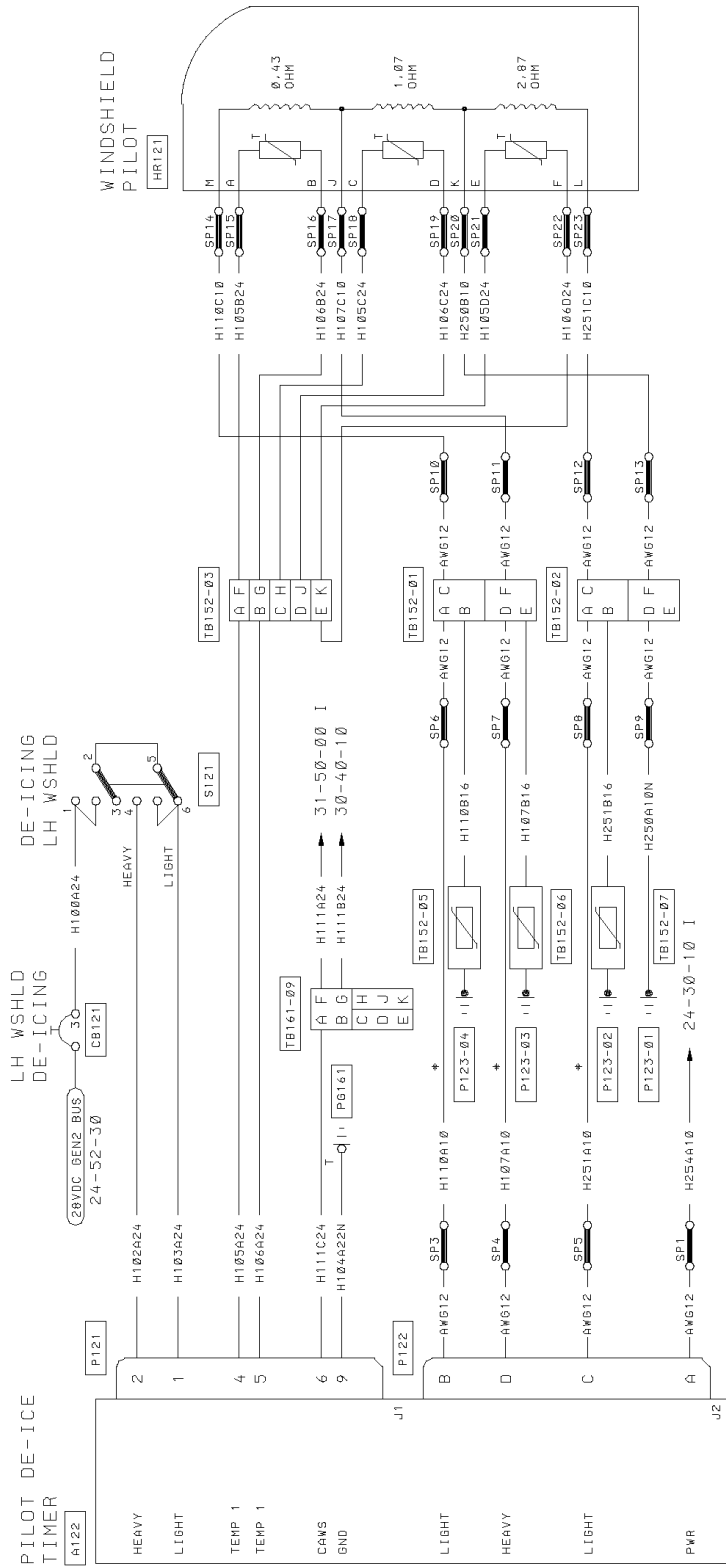
E. Documentation

Make an entry in the Aircraft Logbook that this Service Bulletin has been incorporated.

4. Accomplishment Instructions - Spares

Not applicable.

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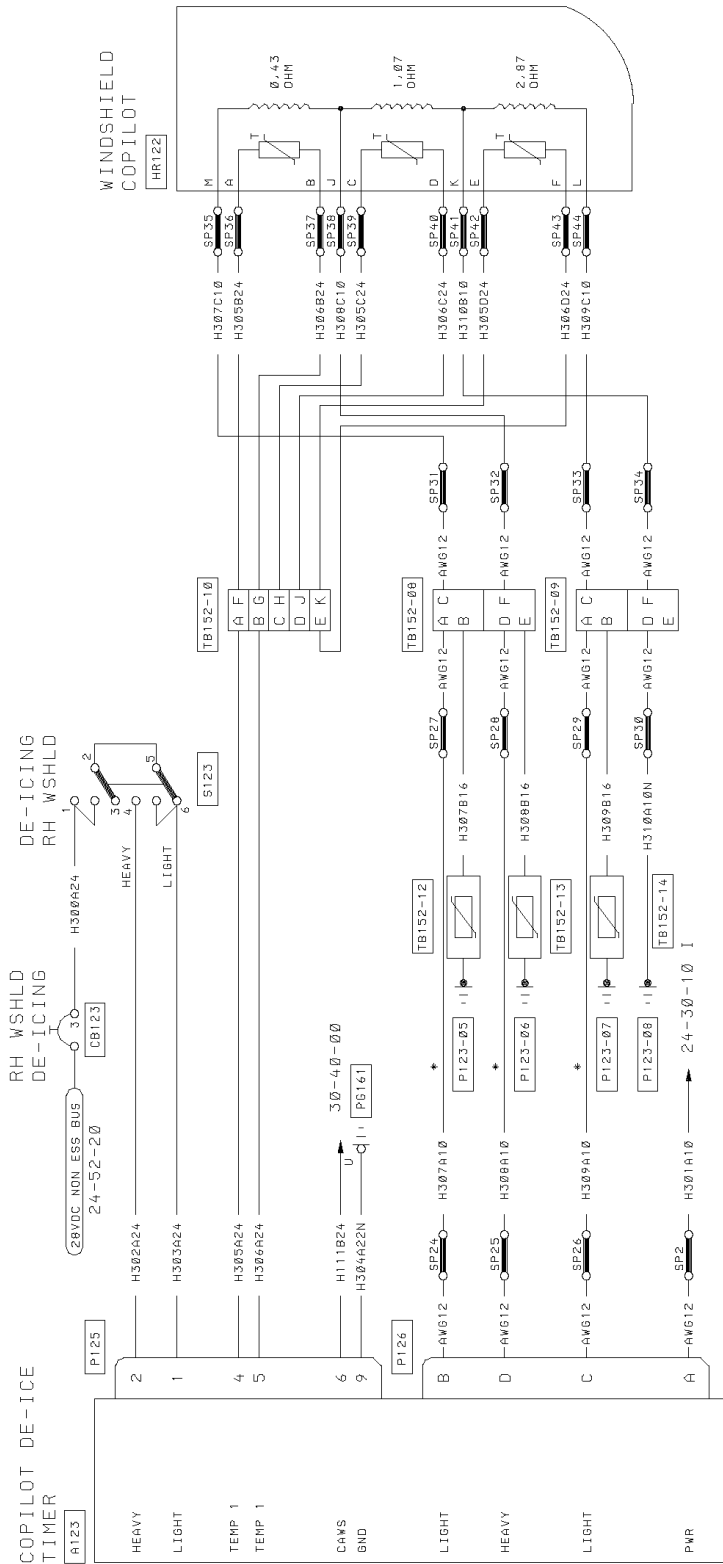
* SEPARATE WIRE BUNDLE

Status / Form. Level	Name	Alt. / Impl.	Version/Status Date Created	Latest Modification Last Modified	Blank Sheet	Total Sheets
CREATED	G. ZAHND	ECA	12.01.00	12.01.00	1	1
LH WINDSHIELD DE-ICING						
File No. 530-49-12-008						
30-40-00						

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Wiring Diagram - Left Windshield Deice System
Figure 1

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* SEPARATE WIRE BUNDLE

Wiring Diagram - Right Windshield Deice System
Figure 2

Status / From / Level	Name	Abt. / Dept.	Uspungungsdatum / Date Created	Letzte Änderung / Last Modified	Blatt / Sheet
CREATED	G. ZAHND	ECA	12.01.00	12.01.00	1 / 1
PILATUS PILATUS AIRCRAFT LTD STANS / SWITZERLAND					PIL-No
RH WINDSHIELD DE-ICING					530.49.12.009
					30-40-10

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